

ABSTRACT

The invention relates to an arrangement for detecting a shaft break in a rotor of a first turbine (10), particularly a medium pressure turbine, of a gas turbine, particularly of an aircraft engine, whereby a second turbine (11), particularly a low pressure turbine, is positioned downstream of the first turbine (10), with an operator element (16) positioned between the rotor of the first turbine (10) and a stator of the second turbine (11) radially inwardly relative to a flow channel, and with a sensor element (21) guided in the stator of the second turbine (11), in order to convert a shaft break, detected by the radially inwardly positioned operator element (16), into an electrical signal and to transmit this electrical signal to a switching element which is positioned radially outwardly relative to the flow channel on a housing of the gas turbine.